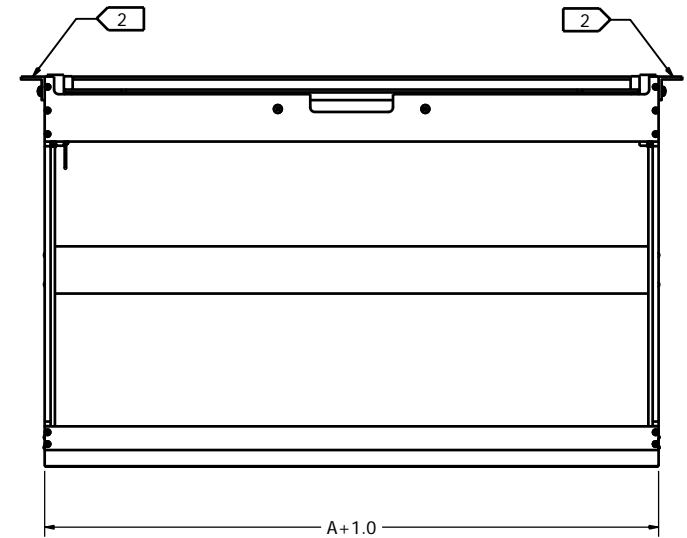
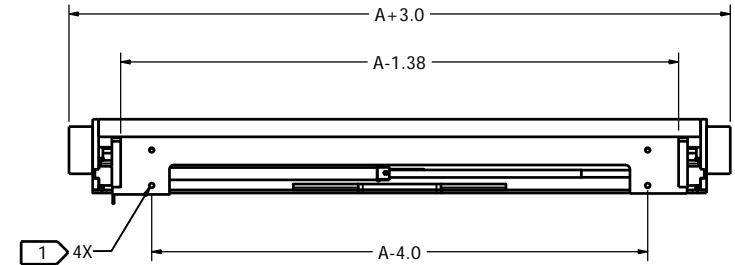
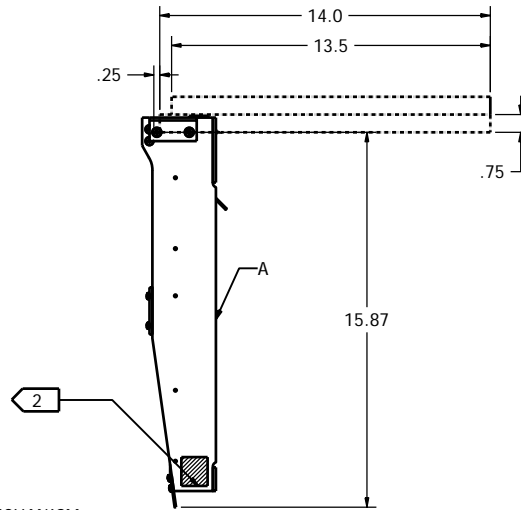
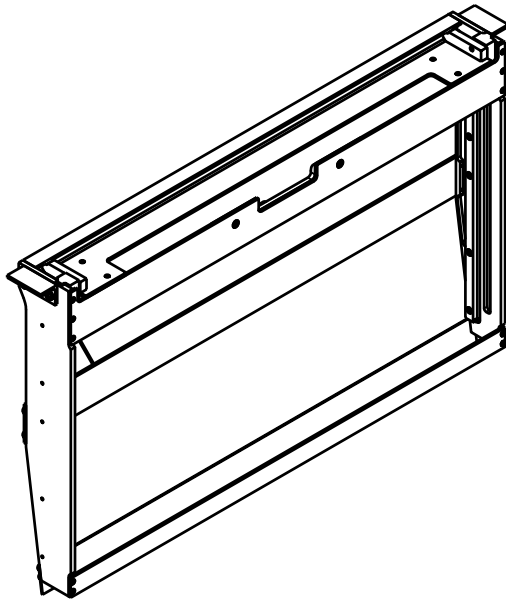


NOTES:

- FOR ADDITIONAL INFORMATION REFER TO OPERATIONAL MANUAL M22180.



**FRONT VIEW**

A = DASH NUMBER (WIDTH OF TABLE)

NOTES:

- ATTACHMENT OF THE TABLE TOP TO THE MODEL 22180 MECHANISM:

- THE TABLE TOP MUST BE FASTENED TO THE MECHANISM WITH A MINIMUM OF 4 NAS514 SCREWS AS SHOW IN THE TOP VIEW.
- THE COMPLIANCE OF THE TABLE TOP TO FEDERAL AVIATION REGULATIONS IS THE RESPONSIBILITY OF THE INSTALLER.

- ATTACHMENT OF THE MODEL 22180 TO THE AIRCRAFT:


- MECHANISM MUST BE FASTENED TO THE AIRCRAFT STRUCTURE WITH A MINIMUM OF 4 NAS514 SCREWS. FASTENERS SHOULD NOT INTERFERE WITH THE OPERATION OF MECHANISM. RECOMMENDED FASTENING AREAS ARE INDICATED.
- STRUCTURAL SUBSTANTIATION OF THE AIRCRAFT STRUCTURE IS THE RESPONSIBILITY OF THE INSTALLER.
- COMPLIANCE TO APPLICABLE FEDERAL AVIATION REGULATIONS REGARDING AISLE CLEARANCE REQUIREMENTS IS THE RESPONSIBILITY OF THE INSTALLER.

3. DESIGN LOADS OF THE MODEL 22180 MECHANISM:

THE MECHANISM CAN SUPPORT 202 LBS. ULTIMATE APPLIED AT 17.2 INCHES FROM SURFACE A.

CONVERT ULTIMATE LOAD TO AN ALLOWABLE TABLETOP WEIGHT AND ITEMS WITH THE FORMULA BELOW:

$$(WEIGHT OF ITEMS + WEIGHT OF TABLE TOP) \times ULTIMATE LOAD FACTOR \leq 202 \text{ LBS.}$$

INSPECTION CLASS C		DRAWN: VVN	DATE: 12/27/2007	 Georgetown, Texas
UNLESS OTHERWISE SPECIFIED		FINISH: NONE		
1) ALL SURFACES PER EPS-121-01-D 2) REMOVE ALL BURRS & SHARP EDGES .010-.015 3) DIMENSIONS ARE IN INCHES 4) DO NOT SCALE DRAWING 5) TOLERANCES:		MATERIAL: SEE NOTE		TITLE: SIDE WALL TABLE 22180 INSTALLATION DRAWING
DECIMAL .XX ± .04 .XXX ± .010 6) ALL MACHINED SURFACES 63/		The data, concepts or designs contained herein are the exclusive property of Enlite, Inc. or contain the proprietary rights of others and shall not be used, disclosed, copied, duplicated or reproduced without the written consent of Enlite, Inc.		SIZE: A DWG NO.: N22180 REV: C SHEET: 1 OF 1